RESEARCH MEMORANDUM

PRELIMINARY RESULTS OF AN INVESTIGATION OF THE EFFECTS OF SPINNER SHAPE ON THE CHARACTERISTICS OF AN NACA D-TYPE COWL BEHIND A THREE-BLADE PROPELLER, INCLUDING THE CHARACTERISTICS OF THE PROPELLER AT NEGATIVE THRUST

By Robert M. Reynolds

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NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

WASHINGTON

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PRELIMINARY RESULTS OF AN INVESTIGATION OF THE EFFECTS
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SUMMARY

An investigation has been conducted to determine the effects of
spinner shape on the pressure-recovery characteristics of an NACA l-series
D-type cowl operating behind a three-blade single-rotation propeller.
Two spinner shapes were considered, an NACA l-series spinner and a modi-
fied spinner, more nearly conical than the l series. The aerodynamic
characteristics of the propeller operating ahead of the cowling and also
operating on the isolated l-series spinner were also determined. The
negative thrust characteristics of the propeller-spinner combination
were measured at low speeds for blade angles from 25° to -20°.

Some of the results of this investigation are being made available
in preliminary form in this report in order to expedite their release.

INTRODUCTION

At the present time there are several new turboprop-powered airplanes
in an advanced stage of development. These airplane designs have the
common problem of providing efficient air induction for the turbine
engine.

It is the purpose of the present report to summarize the results of
tests made to compare the ram-recovery characteristics of a D-type cowl
behind a fairly thick three-blade propeller in combination with an NACA
l-series and a modified-conical spinner. Also, in summary, the maximum
efficiency of the propeller operating ahead of the cowling is compared
with the maximum efficiency of the propeller operating on the isolated
l-series spinner. The low-speed characteristics of the propeller
operating under negative thrust on the isolated 1-series spinner are included.

The present results are offered without discussion in order to permit their release ahead of the more detailed results of the investigation.

SYMBOLS

\( a \) speed of sound, ft/sec

\( b \) blade width, ft

\( c_{\text{r},d} \) blade section design lift coefficient

\( C_p \) power coefficient, \( \frac{P}{\rho n^2 D^5} \)

\( C_T \) thrust coefficient, \( \frac{T}{\rho n^2 D^4} \)

\( D \) propeller diameter, ft

\( \frac{b}{D} \) blade width ratio

\( H \) total pressure, lb/ft\(^2\)

\( \frac{H_1-P_0}{H_0-P_0} \) ram-recovery ratio

\( h \) maximum thickness of blade section, ft

\( \frac{h}{b} \) blade thickness ratio

\( J \) propeller advance ratio, \( \frac{V_o}{nD} \)

\( M_d \) datum Mach number, \( \frac{V}{a} \)

\( n \) propeller rotational speed, rps

\( P \) power, ft-lb/sec
\[ p \quad \text{static pressure, lb/ft}^2 \]

\[ R \quad \text{propeller tip radius, ft} \]

\[ r \quad \text{blade-section radius, ft} \]

\[ T \quad \text{thrust, lb} \]

\[ T_c \quad \text{thrust coefficient, } \frac{T}{\rho V^2 D^2} \]

\[ V \quad \text{datum velocity (wind-tunnel air-stream velocity corrected for solid blockage of cowling but uncorrected for wind-tunnel-wall constraint on the propeller slipstream), ft/sec} \]

\[ V_0 \quad \text{equivalent free-air velocity (datum velocity corrected for wind-tunnel-wall constraint on the propeller slipstream), ft/sec} \]

\[ \frac{V_i}{V} \quad \text{inlet-velocity ratio} \]

\[ \beta \quad \text{propeller blade angle at 0.75R, deg} \]

\[ \beta_d \quad \text{design propeller section blade angle, deg} \]

\[ \eta \quad \text{efficiency, } \frac{TV_0}{P} \text{ or } \frac{C_T}{C_P} \]

\[ \rho \quad \text{mass density of air, slugs/ft}^3 \]

**Subscripts**

\[ o \quad \text{free stream} \]

\[ i \quad \text{location of rake in cowling inlet} \]

\[ a \quad \text{apparent (applied to values of thrust, power, and efficiency measured for the propeller operating in the presence of the cowling)} \]

\[ \text{max} \quad \text{maximum} \]
MODEL AND METHODS

The model of the cowl and propeller was mounted on the 1000-horsepower propeller dynamometer (described in ref. 1) in the test section of the Ames 12-foot pressure wind tunnel as shown in figure 1. Figure 2 is a sketch of the general model arrangement showing the principal model dimensions. The inlet area of the model was scaled (approximately 1/4 size) to match the air requirements of the Pratt-Whitney T-34 engine. Coordinates for the cowling and spinners are given in table I. The three-blade propeller used for this investigation has the designation NACA 3.638-(675)(057)-0572. Blade-form curves for the propeller are presented in figure 3. The platforms between the propeller blades and the spinners were fixed to the spinners at a local blade angle of 83° and had the dimensions and coordinates given in figure 2 and table I, respectively.

The instrumentation of the cowl and the methods used in obtaining ram-recovery ratio and inlet-velocity ratio were the same as described in reference 2, except that for this investigation the total- and static-pressure rakes were made up of six tubes each in place of the eight-tube rakes employed in the tests of reference 2. The methods used in determining the thrust, power, and efficiency of the propeller when operating on an isolated spinner and when operating ahead of a cowling were the same as described in references 1 and 3, respectively. For the propeller operating under negative thrust at the larger negative blade angles, the minimum advance ratio was limited to the corresponding value of \( T_0 \) at which the theoretical effects of the tunnel walls could be confirmed from measurements of wall pressures.

TESTS AND RESULTS

Summary curves showing the ram-recovery ratio as a function of inlet-velocity ratio for the cowl with the two different spinner shapes are presented in figure 4 for Mach numbers from 0.2 to 0.8 with the propeller removed. Similar data are presented in figure 5 for the propeller operating at various combinations of Mach number, blade angle, and advance ratio.

The maximum apparent efficiency of the propeller operating ahead of the D-type cowl is shown as a function of Mach number and blade angle in figure 6. These data are all for an inlet-velocity ratio of approximately 0.5. The maximum efficiency of the propeller with the 1-series spinner with the cowl removed is also shown in this figure. Note that at the higher Mach numbers (0.70 and 0.80) the propeller blade angle is so large that the inner portions of the blade are at local angles greater
than 90°. Stress limitations of the model propeller prevented testing at a higher rotational speed and, thus, the propeller characteristics at lower blade angles could not be measured at these Mach numbers.

The aerodynamic characteristics of the propeller operating under conditions of negative thrust at a forward Mach number of 0.15 are shown in figure 7 as a function of advance ratio and in figure 8 as a function of propeller blade angle.

All data presented herein were obtained at a Reynolds number of 1.0 million per foot, based on the datum velocity.

Ames Aeronautical Laboratory
National Advisory Committee for Aeronautics
Moffett Field, Calif., Oct. 2, 1953

REFERENCES


**TABLE I. - COWLING-SPINNER COORDINATES**

[Coordinates in inches]

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<tr>
<th>Distance from leading edge of cowl, ( x_c )</th>
<th>NACA 1-62.8-070 cowl, radius, ( r_c )</th>
<th>Distance from leading edge of cowl, ( x_1 )</th>
<th>NACA l-series inner lip, radius, ( r_1 )</th>
<th>Distance from leading edge of l-series spinner, ( x_s )</th>
<th>NACA 1-50-74.6 spinner, radius, ( r_s )</th>
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<th>Modified-conical spinner, radius, ( r_{gs} )</th>
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Figure 1.- The model mounted on the 1000-horsepower propeller dynamometer in the 12-foot pressure wind tunnel.

Figure 2.- Model arrangement.

Note: Dimensions shown in inches.
Figure 3.- Plan-form and blade-form curves for the model propeller having the designation NACA 3.636-(675)(057)-0572.
Figure 4.- The effect of inlet-velocity ratio on the average ram-recovery ratio for the cowl with 1-series and modified-conical spinners, propeller removed. Mach numbers from 0.20 to 0.80.
Figure 5.- The effect of inlet-velocity ratio on the average ram-recovery ratio for the cowl with l-series and modified-conical spinners, propeller operating.
(e) $M_d = 0.60; \beta = 53^\circ; J = 2.82$

(f) $M_d = 0.70; \beta = 58.5^\circ; J = 3.40$

(g) $M_d = 0.80; \beta = 58.5^\circ; J = 3.30$

Figure 5.- Concluded.
Figure 6.- The effect of Mach number on the maximum efficiency of the propeller.
Figure 7. - Characteristics of the propeller in negative thrust; $M_d = 0.15$. 

(a) $C_T$ vs. $J$
(b) $C_p$ vs. $J$

Figure 7. Concluded.
Figure 8.- The effect of blade angle on the characteristics of the propeller in negative thrust; \( M_d = 0.15 \).